

Appendix H

LISA Pathfinder thermal stability analysis

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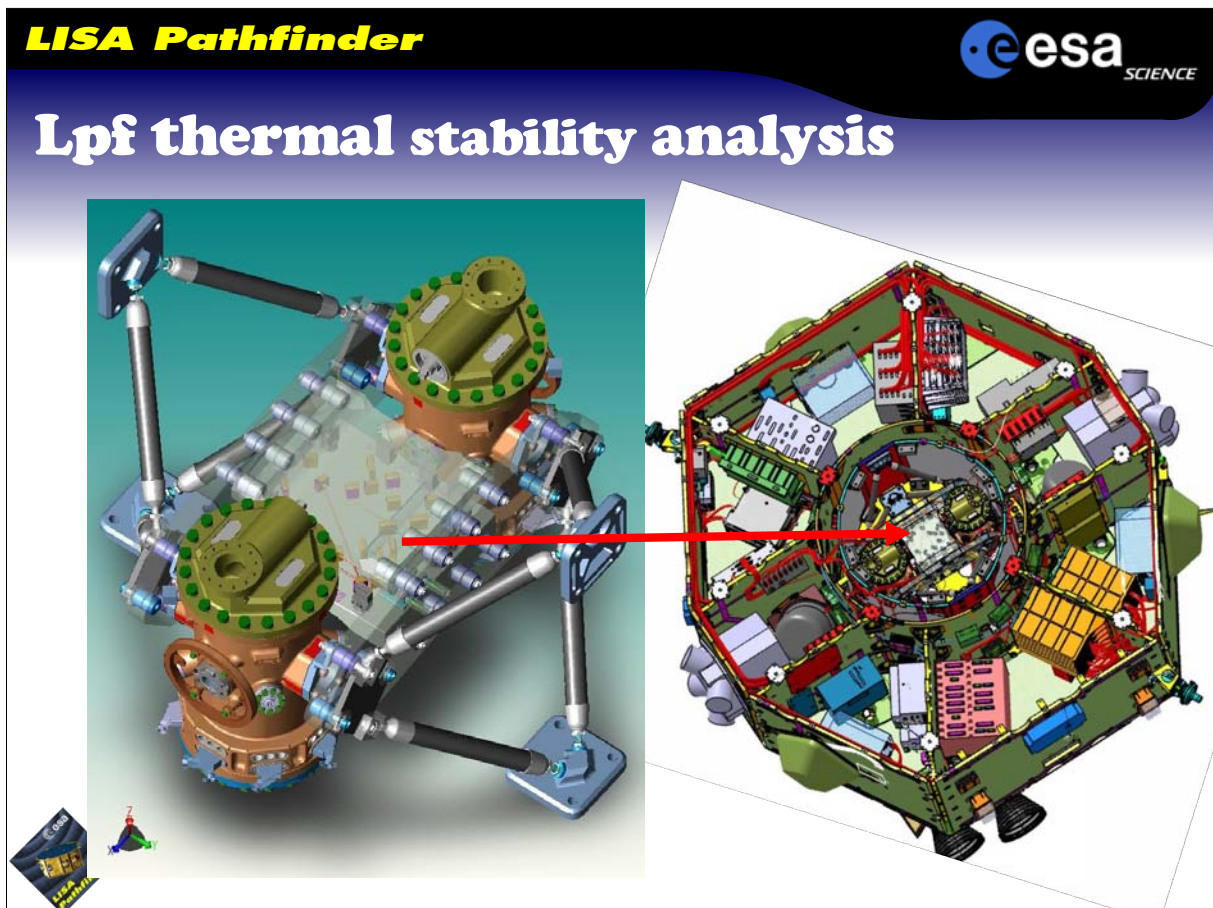
LISA Pathfinder thermal stability analysis

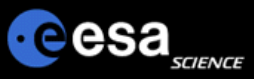
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& ECLS Software

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The slide features a dark blue background with a 3D rendering of the LISA Pathfinder spacecraft in the center. The spacecraft is shown from a perspective that highlights its cylindrical body and various instruments. In the bottom right corner, a small image of the Earth is visible.



LISA Pathfinder 

Lpf thermal stability analysis

The slide displays two 3D CAD models of the LISA Pathfinder spacecraft. On the left is an external view showing the spacecraft's structure, including its boom and instruments. On the right is an internal cutaway view showing the internal components and wiring. A red arrow points from the external view to the internal view, indicating a transition or relationship between the two models. In the bottom left corner, there is a small LISA Pathfinder logo.

LISA Pathfinder















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LISA Pathfinder




LISA Pathfinder

-  First fundamental physics science mission.
-  To prepare for the Lisa mission by testing the concept of gravitational wave detection:
 - Demonstrating that the required force noise floor can be achieved (appropriate S/N ratio will be reached for LISA).
-  To validate in flight technologies not fully testable on ground:
 - Inertial sensor: test mass cannot be free floating on the ground,
 - FEEPs: microNewton force is difficult to measured.



Mission objective

-  Mission objective: verify that a test-mass can be put in pure gravitational free-fall within $3 \times 10^{-14} \text{ m/s}^2/\sqrt{\text{Hz}}$ between 1 mHz and 30 mHz

$$S_a^{1/2}(f) \leq 3 \times 10^{-14} \left[1 + \left(\frac{f}{3 \text{ mHz}} \right)^2 \right] \text{ m s}^{-2} / \sqrt{\text{Hz}}$$

-  Sources of force noise must be minimized:

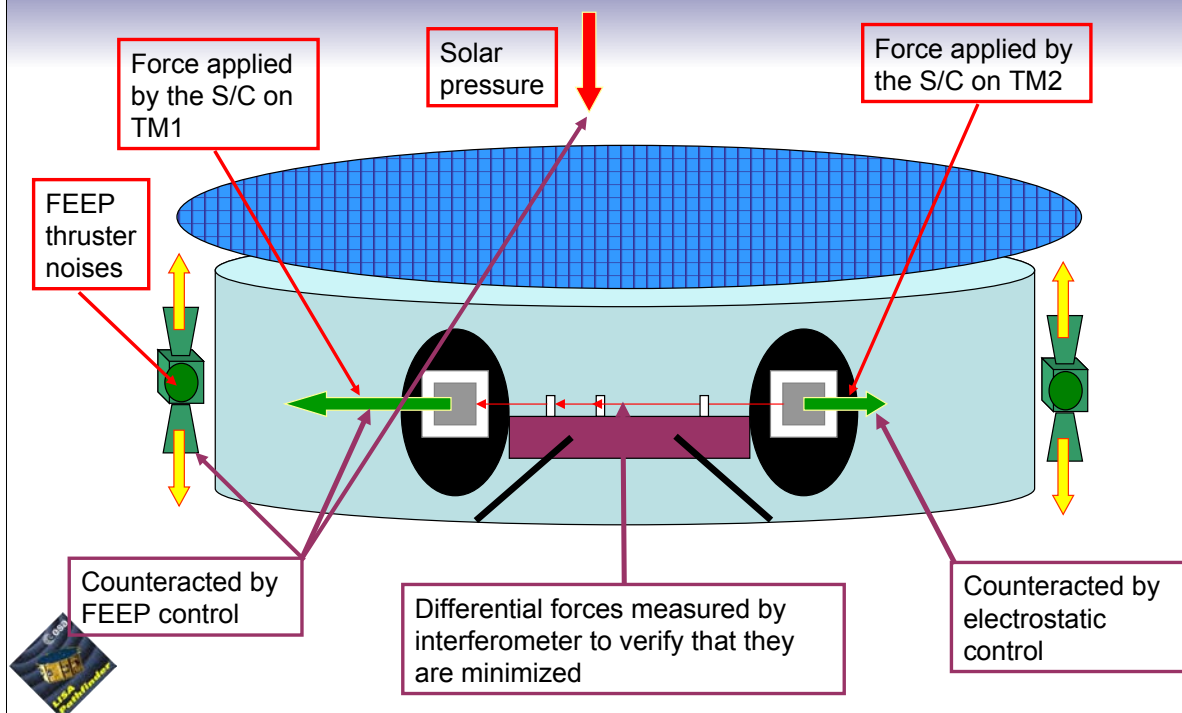
- ⇒ No pressure force or mechanical contact force: “drag-free” mission.
- ⇒ No electromagnetic forces.
- ⇒ No self-gravity forces from SC and instrument itself.

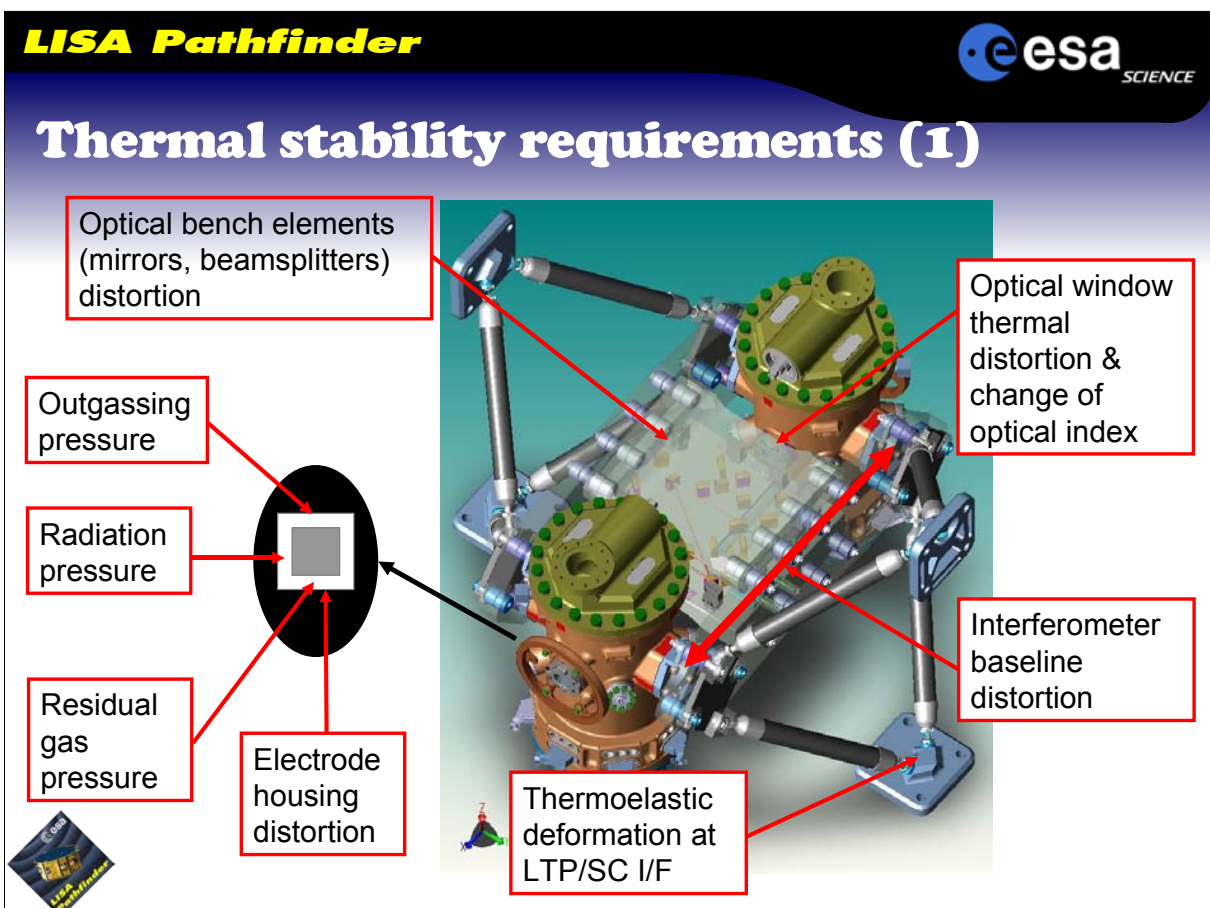
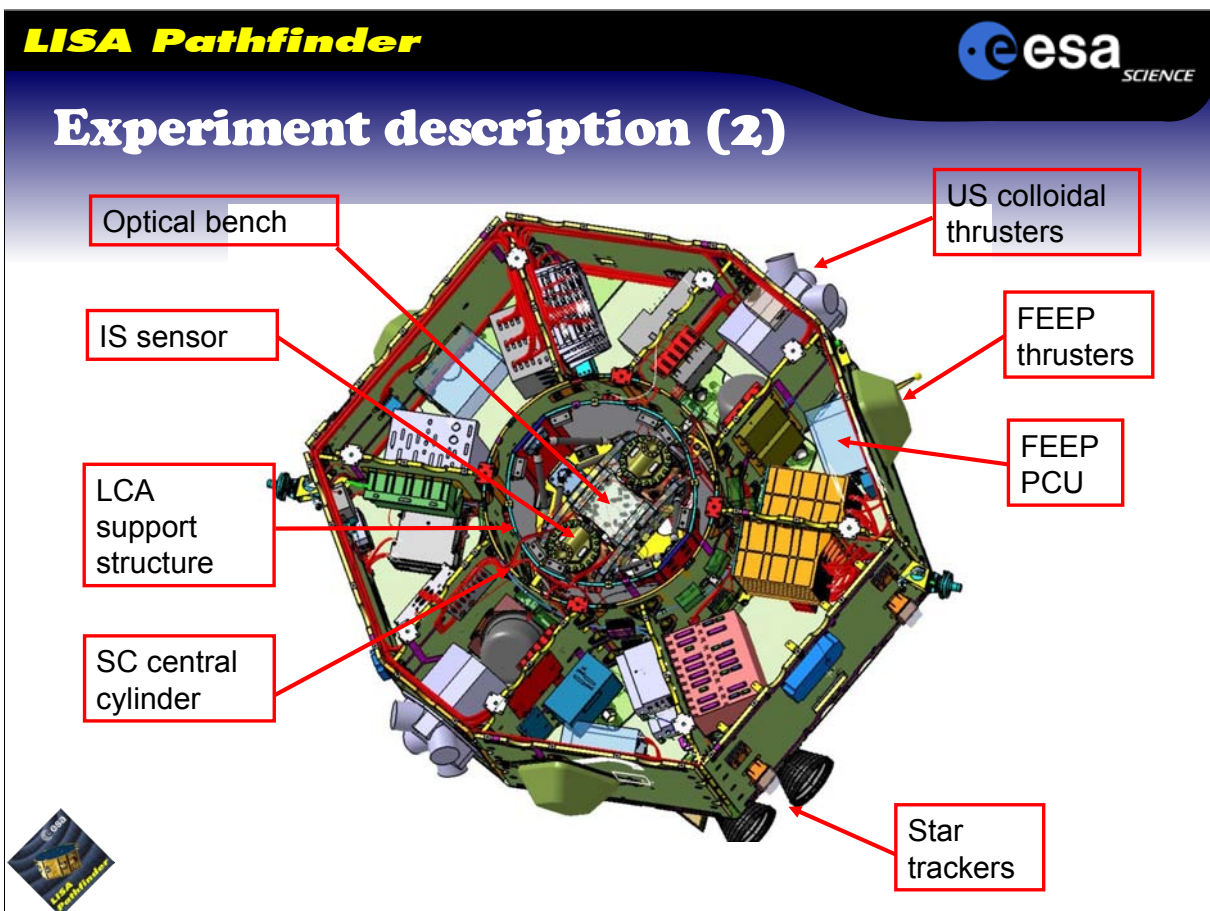
-  Differential acceleration measurement must be one order of magnitude better:

- ⇒ Measurement by interferometry with accuracy better than 9 [pm/√Hz].



Experiment description (1)






Thermal stability requirements (2)

 Mission performance requirements expressed as power spectrum density:

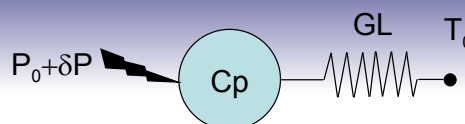
 all thermal requirements are also expressed as power spectrum density.

 What is power spectral density function?


 Power spectral density function (PSD) shows the strength of the variations (energy) as a function of frequency. In other words, it shows at which frequencies variations are strong and at which frequencies variations are weak.




A simple thermal filter (1)



$$C_p \left[\frac{dT_1}{dt} \right] = GL \times (T_0 - T_1) + (P_0 + \delta P)$$

 **Steady-state:** $\left[\frac{dT_1}{dt} \right] = 0$ and $\delta P = 0 \Rightarrow T_1 = T_0 + \frac{P_0}{GL}$


 **Transient:** $C_p \left[\frac{\delta T_1}{dt} \right] + GL \times \delta T_1 = \delta P$

 **Sinusoidal power variation:**

$$\delta P_1(t) = P_{10} \times e^{j\omega t}$$

$$\delta T_1(t) = T_{10} \times e^{j\omega t}$$

$$(C_p j\omega + GL) T_{10} \times e^{j\omega t} = P_{10} \times e^{j\omega t}$$

 $\delta T_1(j\omega) = \frac{\delta P_1}{GL} \times \frac{1}{\left(1 + j\omega \times \frac{C_p}{GL} \right)}$

$\omega=0$



A simple thermal filter (2)

From the linearized thermal model, the thermal filter gain and phase plot can be derived. They express how a power or temperature boundary fluctuation is attenuated at another node by the thermal network.

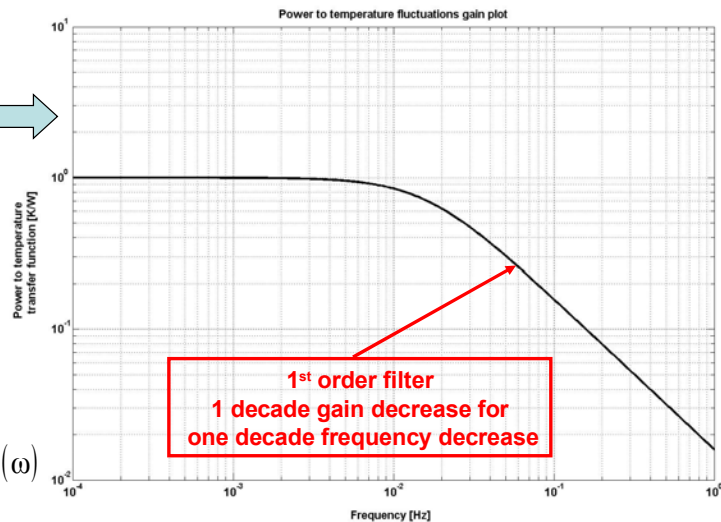
For example for the previous system with $C_p=10$ and $GL=1$.

Power spectrum density is related to transfer function:

$$\delta T(j\omega) = H(j\omega) \times \delta P(j\omega)$$

\Rightarrow

$$PSD_{\delta T}(\omega) = |H(j\omega)|^2 \times PSD_{\delta P}(\omega)$$



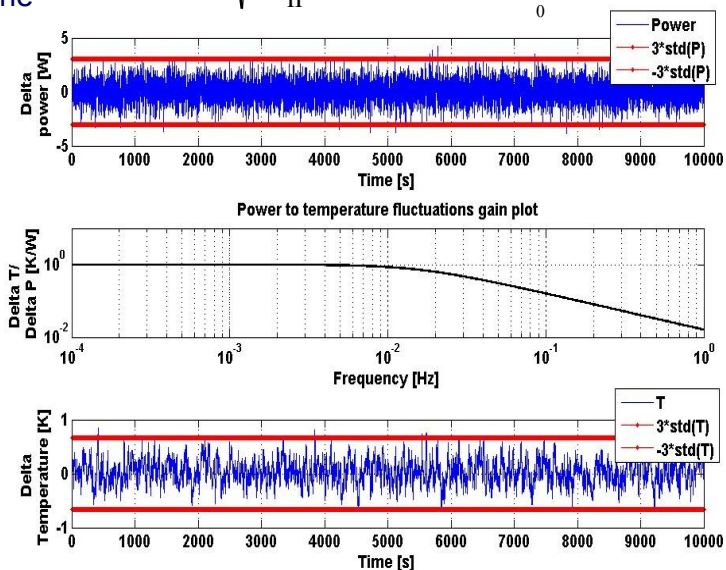
Time and Frequency domain analysis

Root Mean Square (and standard deviation) of input or output signal is related to the integral of the power spectrum density:

$$\delta T_{RMS} = \sqrt{\frac{\sum_{k=1}^n \delta T_k^2}{n}} \quad \text{and} \quad \delta T_{RMS}^2 = \int_0^{f_{MAX}} PSD_{\delta T}(\omega) d\omega$$

If power spectrum density is constant, standard deviation is directly related to the square root of the psd:

$$\delta T_{RMS} = \sqrt{PSD_{\delta T}} \times f_{MAX}$$



Proposed ESATAN add-on

 ESATAN includes the non-linear thermal model:

$$C_{N \times N} \left[\frac{dT}{dt} \right]_{N \times 1} = K_{N \times N} \times [T]_{N \times 1} + F_{N \times N} \times [T^4]_{N \times 1} + B_{N \times q} \times P_{q \times 1}$$

Add-On 1:
Extraction
of matrices

- ☐ C (size N×N) matrix of thermal capacitances of the nodes (diagonal).
- ☐ K (size N×N) matrix of thermal conductances between the nodes.
- ☐ F (size N×N) matrix of radiative couplings between the nodes.
- ☐ P (size N×q) matrix of input power dissipation (and external fluxes).

 Linearize the thermal model: $C \times \left[\frac{d\delta T}{dt} \right] = (K + F \times 4 \times T_e^3) \times \delta T + B \times \delta P$

 Derive the thermal filter transfer function:

$$\delta T(j\omega) = \{Cj\omega - (K + F \times 4 \times T_e^3)\}^{-1} B \times \delta P(j\omega)$$

Add-On 2:
Computation of gain
& phase plots



Thermal disturbances

 During science phase:

- The spacecraft is constantly facing the sun at L1,
- No equipments are switched on or off.

 Major sources of thermal disturbances:

- FEED power control units (3 PCU),
- On-board computer (1 OBC),
- Power Control and Distribution Unit (PCDU),
- Solar flux variation.

 FEED PCU is the leading source of disturbances in the Lisapathfinder measurement bandwidth:

- The applied thermal disturbances can be derived of the control system commands (thrust is going up and down therefore PCU power dissipation goes up and down).



Numerical performance assessment

- Alternative way to determine the thermal stability is to estimate numerically the power spectrum density:
1. Define time series of inputs for FEEP PCU, OBC, PCDU and solar flux fluctuations.
 2. Perform a transient simulation with ESATAN detailed thermal model.
 3. Estimate numerically the power spectrum density of the outputs using fast-Fourier transform techniques.

Problems:

- Simulations needs to be very long without step transient in input power,
- Numerical estimate of power spectrum density is intrinsically limited,
- **Results are difficult to interpret.**

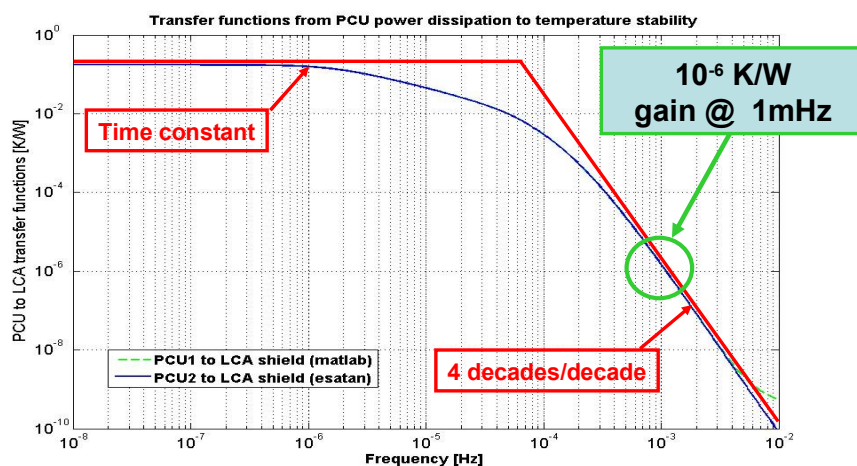
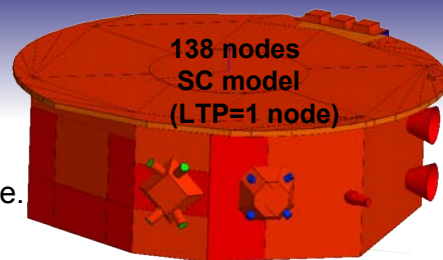


Both analytical and numerical performance needs to be done and compared to provide confidence in the results and enable interpretation.



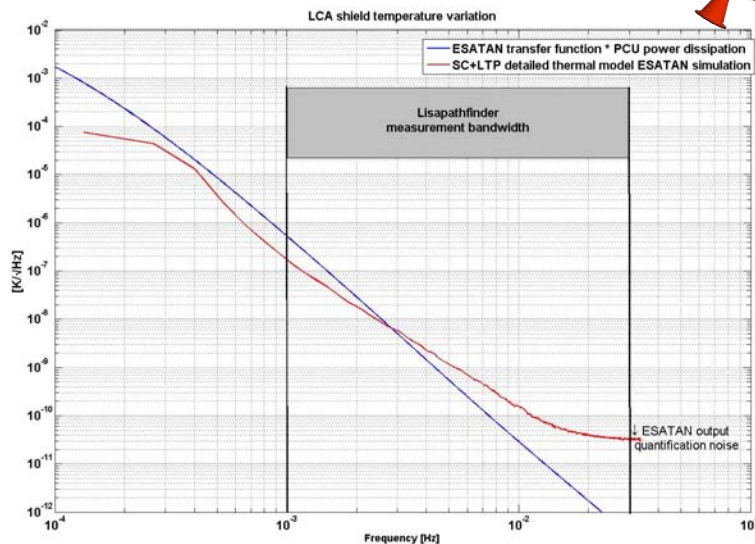
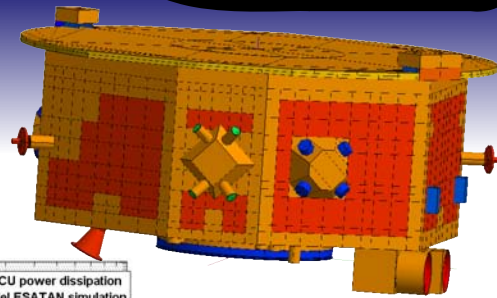
Results (1): analytical

- Performed with PDR reduced S/C model (CLA model without convection):
1. Extracting matrices from ESATAN model,
 2. Using beta version of ESATAN new release.
- Enable only assessment of SC/LCA I/F structure thermal stability.



LISA Pathfinder**Results (2): numerical**

- Performed with post PDR detailed S/C model.
- Enable assessment of all thermal stability requirements.



**SC + LTP model
detailed model
> 6000 nodes**

LISA Pathfinder**Follow-on**

- Evaluate all thermal performance requirements with a reduced LTP+SC model with sufficient discretization for LTP.
- Extract the reduced (condensed) linear dynamic system of thermoelastic behaviour from Nastran and obtain analytically the transfer function between power dissipation and deformation.
- Insert thermoelastic model (and optical model?) into Lpf End To End performance simulator.



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